

5.0 AVIONICS MANAGEMENT

Mr. Herbert E. Grier  
Hon. Willis M. Hawkins



## 5.0 SHUTTLE AVIONICS SYSTEM

### I. BACKGROUND

The Avionics System for the Shuttle is the combination hardware/software system which controls and directs the Shuttle flight. Through its sensors, computers, and interface units it coordinates and implements all functions of the flight except for the specific control of the engine which is done by a separate computer system built onto the engine. The computers of the Avionics system are the nerve center of the Shuttle, and hence must function for the flight to be performed. Appropriate redundancy is built into the system and provision has been made for manual as well as automatic input. The matter of redundancy is not simple, in that the software system itself is a single point failure item except in part for the backup guidance program. This fact is the driver that makes the verification and testing of the software so important in order that the postulated redundancy will be realized.

Because of the criticality of the Avionics System and the inherent challenges in managing this area, the task team meets frequently with the various organizations at the Johnson Space Center and the hardware and software contractors. In addition the team meets with the technical assessment group at JSC and the Chief Engineer to discuss their reviews of this area. Inspection trips are made to both ADL and SDL integration laboratories.

## II. OBSERVATIONS

The current state of the system is that the hardware has been designed and procured. Equipment is coming in and is being de-bugged and operated in the ADL and SAIL laboratories both at Rockwell and at Johnson Space Center. There are hardware and system problems that are being worked diligently and that should be monitored, (e.g., the limitation on Avionics cooling), but the quality of the hardware seems to be very good in light of the stage of the program.

With the hardware in the stage it is in, emphasis has gone to the integration of the various elements and the requirements for their proper operation which, in total, constitute the specification for the software system. There has been an initial design of a software system, but as specific component data become available and mission requirements become more firm, variations or new input must be expected in the software system. These variations are the basis of our concern with the Avionics System.

The computer system in the Shuttle is complicated, and verification of the software is difficult to quantify. In fact, the confidence in software verification is directly proportional to the time spent in such verification; that is, the thoroughness and extent of the verification procedures. In general, one is not confident to say that a software system is reliable unless it has been extensively used.

The criticality of the software system and the difficulty of quantifying its verification make it mandatory to have an independent assessment of the software. Current proposals are to program the testing at ADL and in SAIL so as to perform a complete, independent check of the software. This is a good plan and it must be implemented in a timely manner, and then changes must be rigorously controlled.

The major problem with the Avionics software system is two-fold. First, the tendency of hardware people to solve anomalies in their hardware by changes in the software; and, second, the better definition of the specifications for mission operations which results in a greater software requirement than was initially contemplated for the system. Both of these factors, and particularly late timing, affect the degree of confidence that one has in the formal verification. It is imperative that the computer groups have sufficient time for the software verification, and the simulation laboratories have time to check as deadlines approach. While the first orbital flight is some time away, the ALT flights are almost upon us. The organizational structure to police and drive this program is not readily apparent.

In the course of our discussions several factors became obvious. The first was that the NASA management system is geared to establish communications and coordinate the activities of a number of entities at different locations. However, it does not adequately identify a

specific Avionics responsibility. This system, through its various reviews and panels does, in fact, successfully accomplish a major task of integration, but it is ponderous and time consuming when it must respond to specific, immediate problems in real time. The people in the total system are for the most part very experienced, and an informal system of coping with the real time technical problems has grown up. This system is absolutely vital in that it rings the bells to alert the formal system and supplies the input necessary for the more formal deliberations. This informal system should by no means replace the formal system, but it should be recognized, directed and integrated if the overall structure is to be optimized. From an academic point of view an informal system, with its undefined responsibilities, can sometimes result in balls being dropped, particularly with inexperienced people. We must hasten to say that we feel because of the quality of the personnel the present system is working well. It could perhaps be better defined. We feel that program management recognizes this, that the recent strengthening of the Avionics integration activity will help and that the recognition by the technical assessment group of the importance of the Avionics problem is a good sign. In discussions with the technical people it is quite clear that the integration laboratories (ADL and SAIL), wherein hardware is operated in systems of varying configurations, are

very useful tools. These laboratories provide a real communication channel between all the elements involved in the particular system or subsystem being tested. The joint experience gained here is essential in establishing confidence in the Avionics system and is absolutely necessary as an independent check on the computer software verification.

The whole matter of computer programming and verification is perhaps the element of the system most difficult to assess. The nature of the system and of the current stage of the program inhibits the development of firm computer program requirements. As more simulation experience is generated, for instance, the detailed requirements of manned versus automatic flight undoubtedly will change, resulting in program changes. In addition, the ALT flights will certainly produce data which will require modifications to the programs. As these modifications or new requirements are defined, a continuing effort must be established to police the overall computer program. There is a limit, and there are indications that requirements may exceed the computer capacity. The response to such a situation must not reduce the redundancy built into the computer system.

Verification of a computer program is a subjective and iterative process and it is not easy to assign a confidence number in the same sense that one does with hardware. It is particularly difficult for

the Panel to achieve an assessment in this field. It would be helpful if a single individual were placed in charge.

### III. ASSESSMENT AND RECOMMENDATIONS

The conclusion of reviews to date is that the hardware in the Avionics system is in reasonable shape and that it will perform properly. The software system is currently in a state of flux and is now being given attention, in an effort to scrub down or assign priorities to the requirements and to examine opportunities for simplification. We feel a centralization of control of the software in the program would be beneficial. It is quite clear that because of the reduced requirements on the system for the ALT tests, the load on the computer system is eased. However, confidence in the adequacy of the software, even for this simpler flight program, has still not developed and the Panel must monitor the software program assiduously between the present time and the ALT test.

One conclusion is positive. The Shuttle team, on both the contractor and government side, is composed of experienced, competent people. This fact establishes confidence in the overall program, and assures us that given enough time any contingency can be dealt with properly.

Our recommendations are:

- A. A competent, knowledgeable person should be assigned at the



Program Office level to perform the function of Chief Engineer-Avionics. This may well be the recently appointed Manager, Orbiter Avionics Systems, if he has the central responsibility for the software and the system that it knits together.

B. The program of testing and simulation of the Avionics system should be given a high priority as it forms an independent verification of the software. An additional important benefit of such testing is that it involves a great number of subsystem designers and will form a valuable, real-time communication link in the technical management and integration system.

C. The technical assessment group should establish an appropriate effort to quantify and assess the degree of confidence one can assign to the planned software verification. In our opinion this group should be supplemented by outside experts in the software systems verification field.

D. The recent emphasis on the responsibility of the Avionics Integration Office was a move in the right direction and, if appropriate, further efforts should be made to more clearly define specific software responsibilities.

E. Future actions of the Panel should be limited to monitoring progress of the system so as to judge the state of readiness prior to ALT and the first orbital flight. Should the Panel be expected to assess in detail the software verification, it will need to be supported by an expert in that specific field.



6.0 RISK MANAGEMENT

Dr. Charles D. Harrington  
Mr. Herbert E. Grier  
Mr. Willis M. Hawkins  
Mr. Lee R. Scherer



## 6.0 RISK ASSESSMENT

### I. BACKGROUND

A task team has been formed to review the risk management system and its handling of specific challenges. The task team obtained its information by meetings at JSC and the principal contractor with both managers and the specialists working for them. These meetings were held in September and November 1975, and February and May 1976. Numerous written reports also were provided to substantiate decisions and to demonstrate the procedures used to assure that safety problems are evaluated adequately.

### II. OBSERVATIONS

The areas reviewed included the management system for application of lessons learned from prior programs to Shuttle and the specific cases of the controlled use of teflon insulation, of 26 gauge electrical wiring and of threaded fasteners. The Panel also reviewed the approach to crew and range safety. Finally, we reviewed the approach to assessing and controlling the aggregate or total risk on the program.

#### A. Lessons Learned

The subject of lessons learned is a complicated one. Obviously, a lesson must first be identified as such and there must be agreement as to the proper steps to avoid further occurrence. Once these two steps are properly taken it appears that adequate procedures exist to track the correct application.

Retention methods are:

- a. JSCM 8080 - Standards and Criteria

These are imposed when applicable on subcontractors.

- b. AFSC Handbook DH-1-6

This contains checklists and safety techniques and is used by JSC safety division for checklist inputs.

- c. Various JSC Experience Retention Documents

Examples are:

- 84 Apollo experience retention reports
- JSC 09096 Lessons Learned Skylab
- JSC 0134 B Space Flight Hazards
- JSC 02681 Non Metallic Materials
- JSC 08980 Field Experience Data
- Mission Assessments (Safety), Apollo 7 through ASTP

In addition a lessons learned document has been prepared which states whether the lesson is applicable to Shuttle and how it is to be dispositioned. This document should be continuously updated and safety reviews of Shuttle compared with it. As of June 10th, 1975, the document showed 476 lessons applicable. The question of the proper steps to take to avoid further occurrence is a much more difficult one. For example, the question of man-in-the-loop versus full automation appears to be subject to fine tuning decisions, with some differences of opinion still existing.

## B. Use of Teflon

The use of Teflon is being carefully tracked. It is felt to be the safest insulation material available (where the requirements suggest its use) as long as it is not exposed to temperatures high enough to cause decomposition. There appears, therefore, to be little effort to restrict its use where it is otherwise advantageous. A possible exception is the use inside the oxygen tank of the External Tank. This was originally felt to be safe since only instrument signal current is carried by these wires. However, at the time of the task-team meeting on February 9, 1976, consideration was being given to replacing this section with stainless steel coated, ceramic insulated wiring (as was done in the Apollo oxygen tanks) despite the appreciable weight penalty. Since then the possible acceptability of TFE plastic is being investigated. This reconsideration is occasioned by updated thermal analyses which showed that high temperatures (500°F) may be encountered in use. This item had been closed out in the December 10, 1975, Major Safety Concerns Document (JSC 09990) based upon engineering data and, when appropriate, initiation of new or more extensive engineering analyses. It also illustrates the necessity to maintain a vigilance over revised data and the effect on closed hazards. In this instance, the review system worked when the hazard was reopened.

The cold flow characteristics of Teflon are said not to cause any problems for Shuttle applications. This issue arose during Apollo fabrication days because of a bad batch of Teflon which was

not typical of good quality material. Since then, acceptance tests have been introduced to apply to each new batch of Teflon to assure that no material will be accepted and used in Shuttle which may be deficient in cold flow characteristics. As a result this will no longer be considered a limitation on the places where Teflon may be used. In addition there are firm controls and requirements (Rockwell Space Division Specification ML-0303-0029A and ML-0303-0013 , and Martin Specification STP 6506) which relate to minimum bend radius, clamping force, sharp edges, wire bundle sleeves for protection, harness routing, etc. Rigorous inspection verifies this. Thin walled Teflon has a protective top coat of polyimide resin which restricts cold flow.

C. The Use of Small Gauge Copper Wire

Because of the problem on Apollo with breakage of 26 AWG copper wire the use of this has been largely eliminated, replacing it with 22AWG or heavier. However, in an appreciable percentage of the total footage (2%-8%) it has been found impractical to use wire this large and stiff. Where 26AWG wire has been used it has been made of an alloy of copper having considerably higher tensile strength. It has also been bundled together so that no individual strands can be flexed and broken. OV 101 is being built in this manner. The Panel feels that this problem has been handled properly.



It should be noted that there are many manufacturers' items such as instruments and black boxes which may contain much finer wires. However, these are firmly attached and protected and are not subject to flexing or other mishandling during installation or use. The Panel is satisfied that the design is proper.

D. The Controls on Threaded Fasteners

The Panel found that NASA and its contractors procure fasteners from a variety of sources which meet NASA and DOD specifications. In the manufacture of these fasteners the single element method of gauging is almost always used because it identifies, for the manufacturer, changes in the shape or quality of the threads and alerts the manufacturer to tool and roll wear before the fasteners get out of specification. It is to the manufacturer's economic advantage to use this system since his rejection rate is decreased (i.e., product consistently is of high quality). In addition to gauging, the manufacturer invariably uses an optical comparator and does metallurgical and physical tests on the materials. This whole procedure, statistically applied, insures shipment of high quality fasteners at the minimum price consistent with that quality.

After certification the user, i.e., NASA or its contractors, is primarily concerned with whether a fastener falls within an acceptable envelope of tolerances which can be measured quite rapidly with

go-no go gauges. If the fastener does not meet this test it is returned to the vendor for analysis and replacement. While this might appear to be an arbitrary procedure it is not, because the major factor affecting the failure of a fastener is the proper application of that fastener. Proper application is the facet of the problem that NASA and its contractors must control. Such factors as out of tolerances of parts, insufficient radii at corners, and improper torquing of the fastener more often are responsible for failure than are minor variations in the shape of the thread. We do not believe that one can document a single failure due solely to the threads themselves when they have passed a go-no go inspection. Failures almost always are due to improper application of the fastener and, in a few cases, to a material or metallurgical problem. The improper application of a fastener is prevented first by proper engineering design and review, and second by assembly inspection to see that the proper tolerances are present in the fastened parts and that the correct fastener and torque have been used. The metallurgical aspect of the problem is taken care of by chemical and metallurgical tests as a part of incoming inspection.

The experience of NASA and the DOD, over many years, has resulted in a statistical testing program on fasteners which NASA and its contractors observe. An analysis of these procedures has been

made by NASA and the Panel has reviewed it. In our opinion the program being followed by NASA and its contractors is appropriate and results in the proper degree of safety. We feel that this has been demonstrated by the performance of past NASA projects and by the immense experience of DOD. We further feel that should a fastener failure occur, it almost always will be traced to causes not controlled, or indicated, by the gauging systems.

E. Crew and Range Safety

During launches of the initial Shuttle missions, ground command and destruct capabilities exist on the External Tank and on each SRB. The Orbiter Main Engines cannot be shut down by ground command.

The crew cannot inhibit ground destruct, but are provided warning in advance of such action. Two ejection seats are provided for the crew. Use of ejection seats and of ground destruct devices after the initial missions still is the subject of considerable controversy. There is no precedent in previous programs, since the Shuttle system is a combination of launch vehicle and transport aircraft. Additional complexities result from the split responsibilities between Shuttle program managers and national range commanders, and from the fact that later operational missions will carry "passengers", for whom ejection capability probably would be impracticable.

It is the opinion of the Panel that planning for future missions should proceed with a fundamental ground rule that the capability for destruct by range safety personnel and the capability of escape by all people onboard go hand-in-hand.

Under current plans, adherence to this ground rule would mean that both ejection seats and destruct systems will be removed when more than two people are on board. It seems reasonable that removal of such devices will be an acceptable risk after demonstration by a few successful flights.

F. Response to Recommendations on Hydraulic Fluid

The Panel earlier had recommended that the choice of hydraulic fluid be re-examined.

On November 18, 1975, detailed presentations were made on the comparison of Yellow Oil (MIL-H-83282) and Red Oil (MIL-H-5606) for use as hydraulic fluids. These comparisons showed that Yellow Oil appeared superior to Red Oil in regard to flammability over a narrow temperature range and under certain physical conditions. In some other respects, such as corrosion and low temperature viscosity, Red Oil was superior. The decision has been made to stay with Yellow Oil due to its lesser fire risk. Precautions will need to be taken to keep out water (corrosion) and to avoid excessively low temperatures.

#### G. The Risk Management System and Aggregate Risk Assessment

The Panel found a well-developed independent hazard identification and risk assessment system, the members of which participate in program decision making. They provide formal reports to program management such as summaries of major safety concerns and of the actions being taken to assure management awareness. They have also just completed the initial mission safety assessment report for the ALT flights.

The Panel gave particular attention to management control of both the total or aggregate risk on the program as well as the control of specific hazards.

Aggregate risk has been defined by the JSC Safety Division as the sum of the effects of hardware and operational hazards upon the event, series of events, or mission, and is measured in terms of adverse impact on personnel or critical equipment. The management approach to this assessment is through the safety concerns procedure. In this procedure all inputs to safety questions, including RID's are examined through System Level Hazard Analysis, in preparing the Shuttle level SAR, and screened by a Criteria Committee. They are either resolved through modifications or accepted as risks. They become part of the Safety Concerns Index and Safety Concerns Summary Report and as such are direct input to the Mission Safety Assessment. The latter becomes the true evaluation point for aggregate risk assessment. It

appears that this procedure is adequate from a management point of view to assure that all safety issues, once identified as such, are properly tracked and assessed.

While major hazards are brought before management for their evaluation there is also the question of how you control minor risks and evaluate their impact on the level of aggregate risk being accepted in the program. This is no simple matter because management cannot review every decision and there are not the resources to work every "what if" situation. Therefore, the task team has been in discussions with the safety offices on how to strengthen controls or audits in this area. As a result additional controls have been instituted.

The Screening Board for the "Major Safety Concerns Document" has been passing judgment only upon those issues which are considered significant safety drivers and hence has not reviewed those having little impact. To perform a check of the disposition of these minor risks, the Screening Board has instituted a new procedure whereby it will include an audit of twenty minor issues at each Screening Board meeting to determine that they have been properly evaluated and dispositioned. If the audit reveals deficiencies, a more extensive investigation will be completed. It should be noted that Board membership has been recently revised to include KSC and MSFC representation. The method of assessing the total impact of

these risks is to track the safety issues for satisfactory closeout and to report on them in the Mission Safety Assessment Documents. These documents contain the Safety office's judgment on the acceptability of the "aggregate risk." This is a subjective, rather than quantitative, evaluation of the cumulative accepted risks and actions being taken to resolve open items.

The Panel met with senior program management to review their approach in developing policies that determine the criteria for risk assessment and decision making at subordinate levels. These discussions also included senior management's approach to decision making at their level where it has been their judgment to accept risks. The Panel was both reviewing critical decisions that have already been made and reinforcing management's controls to assure that safety not slip from its normal top priority because of cost and schedule pressures in the period ahead. Among the points made by management in these discussions were:

1. Decisions involving any significant reduction in program requirements are reviewed by senior management to assure a judgment that is objective and sensitive to the requirements of public accountability. This is evidenced by the way the decision was made on contingency abort capability during the SRB burn period.

2. Any decision on safety is a judgment on how far

to go to enhance or guarantee safety. There are specific areas where safety margins have been reduced but the management judgment is that the margins are still sufficient.

3. Redundancy is not synonymous with safety because the complexity of a redundant system may introduce new hazards that reduce the overall safety of the system. Excess redundancy, or appended protection systems, may cause engineers to produce designs that are not optimum but depend upon these additions to make them acceptable.

4. The number of single failure points that could cause critical situations are not greater than in Apollo or Skylab. In fact, Shuttle has a higher safety factor because of the flexibility available to terminate the mission.

5. Aggregate risk is hard to measure but the program is making a conscious effort to identify the magnitude. The Mission Safety Assessment document is one judgment. The program SR&QA people are preparing a form of aggregate risk assessment associated with the program requirements review results.

6. The ground test program provides the best assurance that we understand the system, its capabilities and limitations. While some changes have been made in the test program, piggybacking tests or deferring them, basic requirements have not been compromised.

7. The ALT flights and the subsequent orbital flight



program will develop confidence in the vehicle. They provide for moving into situations of greater risks in carefully considered increments, so that the new risk on any one flight is acceptable or cannot reasonably be reduced further.

### III. ASSESSMENT AND RECOMMENDATIONS

The Panel's judgment as to whether the total aggregate risk is acceptable can only be arrived at over the course of time after careful study of the mission assessment documents and other pertinent data. Once the program is beyond the development flights and is in the operational phase, aggregate risk should be minimized by experience and by the repetitive nature of the flights. Safety questions which the Panel considers significant are being worked, although the resources available may not permit in-depth investigation of all minor issues.

The concept of re-usability introduces a new type of risk in the Shuttle program which was not encountered in previous, single-shot programs. For example, the TPS and the landing requirements introduce a number of safety problems for which experience is lacking.

The final aggregate risk assessment should focus heavily on "what if" questions.



7.0 GROUND TEST AND GSE PROGRAMS

Lt. Gen. Warren D. Johnson, USAF



## 7.1 GROUND TESTS

### I. BACKGROUND

The Aerospace Safety Advisory Panel has studied NASA philosophy pertaining to the entire Space Shuttle System, the "Space Shuttle Verification Program" and particularly the ground tests aspects of that Verification Program. Since the Panel has been in existence for several years and was involved in Apollo, Skylab, and the recent joint US-Soviet Apollo-Soyuz space flight, an inevitable comparison with these programs is made and, indeed, the uniform approach to testing reflects NASA experience. Past NASA programs have been eminently successful. Yet even NASA has suffered temporary failures, and the Panel was created as a result of a disastrous accident. The Panel is conscious that NASA faces a need for major cost reductions in order to stay within programmed costs for the Space Shuttle program. This cost reduction effort could impact on safety unless management review is thorough. A part of our examination focused on this possibility.

The Panel is examining the Ground Test Program as it pertains to preparation for the Approach and Landing Tests, to the Orbital Flight Tests and eventually the operational orbital flights. Activity to date has concentrated on the pre-operational phases. The major effort has been to assist NASA in assuring the Space Shuttle System will fly safely as a space vehicle and as an aircraft when it

reenters the atmosphere to return for landing. In gathering data we have studied the planned Space Shuttle Verification Program, some individual ground tests, and the Hawkins Review to identify possible problem areas. Based on those studies, visits to Rockwell and the Johnson Space Center have been made.

As previously indicated the Space Shuttle Verification Program, and specifically the ground test portion, is based on past highly successful NASA programs. Experienced NASA management has designed and tracked the program since the go-ahead for Space Shuttle was given in 1969. There is a strong reliance on this past experience and an excellent use of "lessons learned." However, ~~ma~~ major NASA programs in the past have dealt with Space Vehicles, one time flights, and better funding priorities. Moreover, past programs were experimental in nature as opposed to operational. Thus, new problems can be expected.

The Ground Test Program is extensive. Obviously, the Panel cannot examine all details, nor is that desirable or necessary. The Panel's contribution should be to identify areas in which there are risks not faced in past NASA programs and/or areas in which previous difficulties have been encountered. Activities to date have identified these priority areas for Panel examination.

## II. OBSERVATIONS

The Ground Test Organization appears adequate. The Test Organization is sufficiently distinct from the organization which designed the Shuttle. Thus, testing objectivity should be assured.

It also appears that there is a reasonable mix of space vehicle and aircraft experience. Rockwell is applying its considerable aircraft expertise to the Space Shuttle Systems, as well as its space experience. They realize the Orbiter must perform as a space vehicle and an aircraft. NASA has an adequate mix of Space experts and pilots who have flown and tested aircraft, including "lifting bodies" with shuttle-like characteristics. The astronauts are deeply involved in the planning and the ground test programs. Throughout NASA there is a reasonable balance of scientists, engineers, engineer-pilots, and other skills. Cost reduction efforts and ensuing personnel reductions have, as yet, not destroyed this core of capability.

An adequate interface between Rockwell and subcontractors appears to exist. The Rockwell organization indicated a realization of the responsibility for monitoring tests conducted by subcontractors. Any test failure must be reported within 24 hours and Rockwell monitors compliance. This will be further checked by the Panel in visits to subcontractors.

Because of funding constraints, some tests have been cancelled.

It appears, however, that management has provided an adequate review of the risks involved in each such reduction.

### III. ASSESSMENT AND RECOMMENDATIONS

The Ground Test Program as originally envisioned had a larger scope of full scale model tests. In the reduction a greater reliance was placed on quarter (1/4) scale model tests. Additional cost reduction efforts have led to some modification of 1/4 scale model tests. Also, some originally scheduled test conditions changed due to lack of availability of components. Planned full scale model tests were directly related to 1/4 scale model tests - designed to provide a one-to-one comparison in such areas as Influence Coefficient and Stiffness Characteristics. The lack of these one-to-one comparisons could have an adverse impact. Management is aware of these reductions and has assessed the risk.

The Panel was concerned with the adequacy of structural testing prior to ALT and has inquired into this at some length.

A. Structural testing of the Orbiter was compared to the testing of the Boeing 747, the Douglas DC-10 and the Lockheed 1011 (similar wide body aircraft). The two former were tested to a greater extent. The 1011 testing was more limited and would tend to indicate that the Orbiter test plan is adequate.

B. ALT will not include thermal and ascent stresses which will



be encountered in orbital flights. Structural analysis prior to ALT assumes these stresses are present, thus creating a margin of safety. However, actual structural tests will not be completed prior to ALT.

C. The Orbiter will be limited to 75% of structural loads (limiting weight and G-forces), during the ALT. The extent of ground tests in this respect is somewhat less than that to which wide body aircraft have been subjected prior to first flight. Perhaps requirements for wide body aircraft are not appropriate for Shuttle. On the other hand, even higher standards might be appropriate. It is suggested that this be a subject for a later meeting of the entire Panel.

There is concern about the testing for the Payload Bay Doors. It is clear that failure to close these doors would preclude safe reentry. Many steps are being taken:

A. NASA (JSC) is making a comprehensive study of the history of "jams."

B. Conservative "overreach" is planned.

C. Many tests are planned.

D. EVA capability is being planned. Tools are being considered and an EVA working group exists.

1. However, some payloads could preclude access by EVA.

2. There is some indication that test payloads during early Orbital Flight Tests are being considered that could interfere

with manual back-up for closing payload bay doors. Recommend no such payloads be permitted during early OFT.

No schedule margin exists in the event any major problems are encountered in ground testing. This is a success-oriented program and any major problems will impact dollars and schedules. This could induce shortcuts that have safety implications. The Panel should examine any major test failure and/or change in the test program in order to act as an additional safeguard to the normal NASA management review.

The review of changes and deletions to the Ground Test Program appears to have been adequate to date. Further budget constraints or a major problem could induce more changes. The Panel believes the "point of diminishing return" must be close for changes in the Ground Test Program. Thus, such changes should be brought to the attention of the Panel as soon as they are defined.

## 7.2 GROUND SUPPORT EQUIPMENT

### I. BACKGROUND

Planning for and acquisition of Ground Support Equipment are largely management problems as opposed to safety issues. However, the Panel notes that such equipment acquisition for various past programs traditionally has been the first to suffer in budget cuts. Moreover, planning is difficult in the early stages of a program, pending development of a firm maintenance baseline. Thus when cuts or changes are made, little time remains to adjust, and equipment deliveries often lag operational requirements. Some safety impact may then result, especially when ground handling and turn around are so dependent on specialized and sophisticated equipment.

The planned turn around of 160 hours would be made more difficult to attain if equipment were not available in the configuration and numbers required.

Orbital Flight Tests could be hampered if Ground Support Equipment were not available. Delays in flight tests could be costly and/or could impact on safety if shortcuts are attempted.

It appears prudent to examine whether the pressure to achieve the 160 hour turn around could create safety problems.

If inherent safety problems exist in the interface between Ground Support Equipment and flight hardware, the Panel wishes to identify them and assure itself these hazards are given adequate attention.

## II. OBSERVATIONS

JSC and KSC are aware of the criticality of Ground Support Equipment and of their responsibility for integration. Both are developing detailed planning for such equipment, considering life cycle requirements and hazard analyses across the interface with flight hardware. Both centers are working closely with the Air Force, which eventually will operate the Space Shuttle System from Vandenberg. Air Force personnel are on hand at JSC and KSC for this purpose.

All seem to be aware that the 160 hour turn around forces better planning for support equipment. However, they assert that they are guarding against the possibility that the turn around requirement could influence shortcuts. They clearly state that the 160 hour turn around is a goal for the operational phase and that it will not be attempted in the orbital flight tests or in early operational flights.

Planning is tied to vendor (subcontractor) availability. If a vendor's production line is planned to be closed or reduced, JSC plans to review the need to acquire support equipment prior to any such action.

Most testing during Orbital Flight Test and in later operational flights is planned to be accomplished on-board the Orbiter, as distinguished from bench checks in a separate facility. Before attempting to repair a black box the malfunction will be clearly identified.

### III. ASSESSMENT AND RECOMMENDATIONS

The Panel should continue surveillance of Ground Support Equipment and should examine the interface of some of the more critical items with flight hardware.

Panel interest should focus initially on equipment required for auto land tests. (Subcontractor equipment is planned to be used to cover most requirements for this and Orbital Flight Tests.)

The Panel also should follow changes and/or reductions planned for support equipment, assuring that NASA reviews of such actions consider all risks involved. (The NASA review process should equal that for changes in the ground testing program.)

The Panel should question planning for Ground Support Equipment as it visits selected vendors (subcontractors) and NASA centers.



8.0 FLIGHT TEST PROGRAM

Mr. John L. Kuranz  
Mr. Lee R. Scherer  
Lt. Gen. Warren D. Johnson, USAF





## 8.0 FLIGHT TEST PROGRAM

### I. BACKGROUND

The Panel undertook to study the Approach and Landing Test Project for the purpose of assessing the value and risks, in order to determine if programming and/or management system changes should be recommended to meet the primary test objectives. We believe these objectives to be valid; they are:

- A. To verify operational capability of the mated ferry configuration.
- B. To confirm the subsonic aerodynamic characteristics of the Orbiter and verify piloted and automatic approach and landing concepts.
- C. To correlate wind tunnel data and flight data. An integral part of the Panel's study was the examination of potentially hazardous conditions associated with the design or operation of both the flight and ground systems.

The Panel's most recent meeting with ALT management was May 24-25, 1976 at JSC. This was preceded by the following activities:

- A. Met with ALT and Carrier Aircraft project officers at JSC on November 18-19, 1975. Detailed discussions on the 747, Orbiter 101, mated configurations and most current test and analytical data supporting the ALT requirements and management decisions.
- B. Session with ALT project personnel at Rockwell International at Downey, California on October 29, 1975. Discussions related to

Rockwell International's participation and implementation of their role in the ALT project.

C. Shorter but significant fact-finding sessions were conducted in Washington at NASA Headquarters on August 28, 1975 and at KSC on December 3, 1975. These served to provide an overview of the ALT project and indicated where further examination would be fruitful.

D. Attendance at the Orbiter ALT Critical Design Review conducted at JSC on April 21, 1976.

E. Panel review and task team sessions at JSC, February 9-10, 1976.

These activities served to provide a well detailed and up-dated background for further fact-finding and gave an integrated perspective to the Panel. Included were major achievements that contribute to program management's confidence in achievement of ALT objectives.

In addition to these face-to-face sessions, numerous program documents were supplied, including the ALT Project Management Plan which, together with the candid and helpful dialogue with program managers and engineers, allows the observations and assessments which follow.

Before reading the section of this report covering observations and assessment, it is worthwhile to review the ALT Project background. ALT covers only a small portion of the Shuttle Verification Program.

Orbiter 101 and a modified Boeing 747 will be used for these tests. Orbiter 101 configuration will be oriented toward the subsystems required for subsonic atmospheric flight. For the most part it will not include subsystems required for space operations. Although not carrying actual payloads, the Orbiter 101 will employ simulated payload structure adequate to demonstrate the effects of payload weight, center-of-gravity, and inertia on approach-and-landing performance. The ALT project includes vehicle ground tests before the first drop flight, preliminary flight evaluation, flying quality investigation of the launch combination, the separation and the Shuttle subsystem verification, and demonstration of the unpowered approach and landing.

## II. OBSERVATIONS

The Shuttle program by nature of costs and schedule constraints is a success-oriented program. This is exemplified by the assignment of a single Orbiter and a single carrier aircraft to this program and the use of the carrier for all future ferry-type operations. Major schedule perturbation would result from mishaps or system failures which could occur during the ALT process. The goals of the program appear to be proper, however, and the tight planning does not at this time imply any increase of risk to the crew during this test series, in ferry operations or in the orbital flight tests that follow the ALT.

It appears that the flight performance data and overall experience to be gained during the ALT activities as currently planned do justify performing this series of tests. This viewpoint is based on an assessment of the risk of performing the ALT versus the risk in eliminating it. While the Panel believes that no single flight test requirement for ALT would in itself justify the program, we believe that it is justified by the aggregate results.

The continuing effort of Shuttle management to utilize the ALT project to its fullest has been a forcing function in establishing details of the ALT. For example, the configuration of the hardware and software is such that it will have the capability of meeting alternate configuration options, tailcone on, tailcone off, etc., depending upon the results of the first few captive and free-flight tests.

Current plans now call for five tailcone on and three tailcone off free flights in addition to the original captive inert and inactive flights. The use of the tailcone on the Orbiter is the result of wind tunnel tests and detailed analyses which show a high degree of 747 tail buffet with tailcone off as the Orbiter is being carried on top of the 747. Significant effects of this buffeting are:

A. Fatigue of the 747 tail area P. However, based on wind tunnel tests and analyses, the structural capability will not be exceeded.

B. The possibility that the mated configuration buffeting will adversely affect flight control, as well as the 747 crew's ability to accomplish required maneuvers.

The ALT management system was discussed in some detail with both the NASA and contractor personnel during the fact-finding sessions. It appeared that the management system, including the reviews and information flow, has been effective in supporting the ALT project; however, there was some indication that not all current information had been communicated on a timely basis. The ALT CDR identified this problem and adequate steps are being taken.

### III. ASSESSMENT AND RECOMMENDATIONS

A. The Panel agrees that an adequate Approach and Landing Test Project is necessary to the orderly and safe development of the Orbiter, the ferry utilization, and other aspects of the overall Shuttle program, both ground and flight.

B. The information gained from the ALT is important to the confidence level required in making the first manned orbital flight with the full Space Shuttle system. The value of the ALT project though, is wholly dependent upon the results of each individual step within the project. A willingness to alter the test program flights as data is collected is expected, which will enhance the synergistic results from all tests.

C. As an aerodynamic vehicle, the Shuttle aircraft is new in many ways. It may exhibit some characteristics in various flight conditions that are not accurately predictable from wind tunnel or other data. The Panel believes that the flight control system, if provided with a cockpit gain variation, would add to the safety of the first flight tests of the Orbiter vehicle. The Panel is aware that the ALT CDR considered this problem; however, we suggest further review.

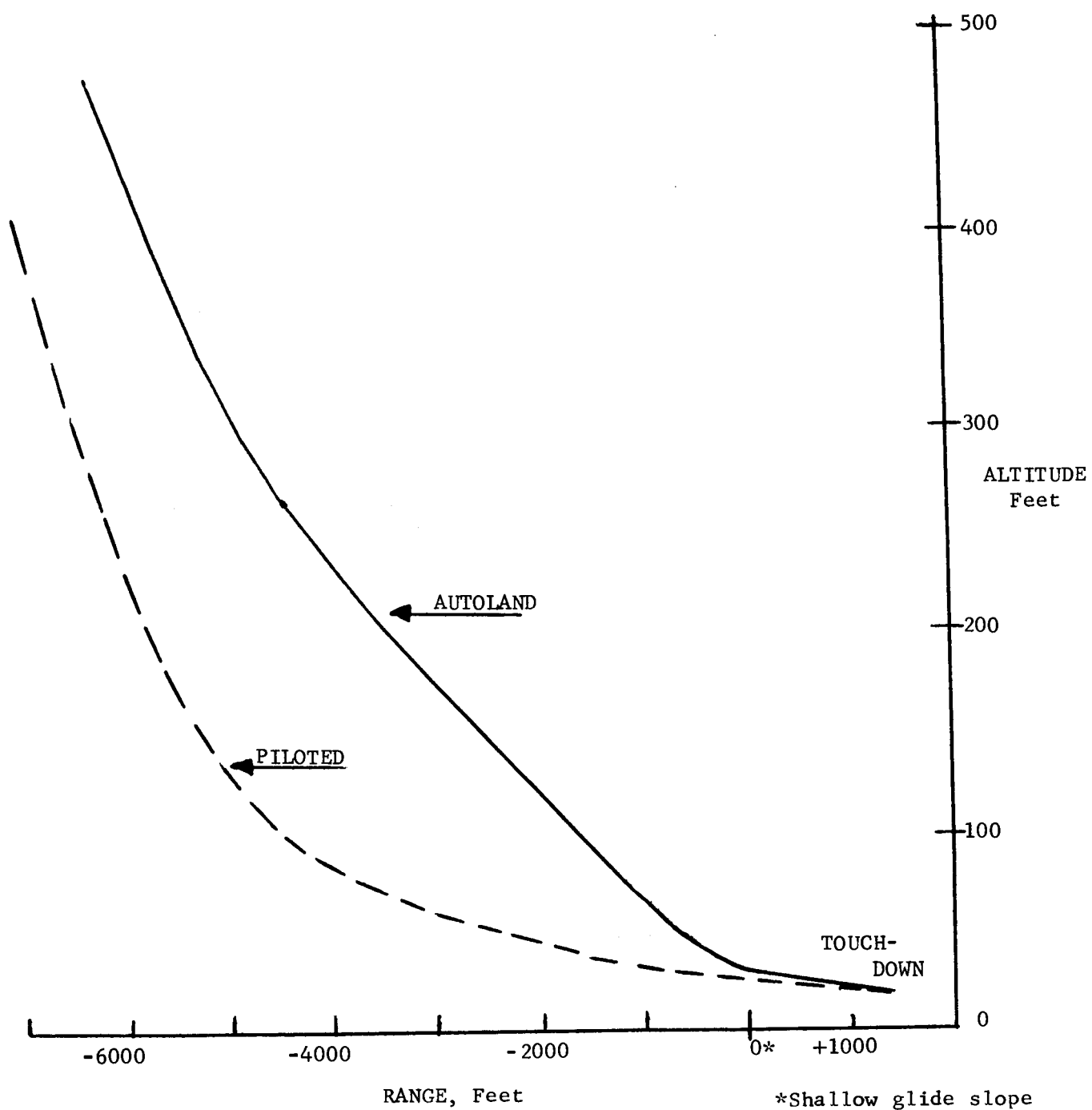
D. If the Orbiter L/D is to be simulated when it is flown with tailcone on, the Panel recommends that extra caution be employed to assure there is sufficient attitude control available when drag devices are deployed. It is realized that currently such maneuvers are not planned.

E. The profile or energy management for approach, flare and landing are different for autoland and manual control modes. Figure 1 shows this difference. Effort is now underway to make the automatic and manual profiles identical. The Panel believes this to be essential. This will make it possible for the crew to follow the progress of an automatic landing, and, if necessary, accomplish the transition from automatic to manual with a minimum of exposure to error.

F. Lifting body flight tests show that successful unpowered landings are best achieved following float profiles that are much

flatter than is now planned for ALT. The Panel recommends further review of the planning and training for the float segment of the ALT.

FIGURE 1 - Altitude versus Range (Typical trajectory)





9.0 EXTERNAL TANK

Dr. William A. Mrazek



## 9.0 EXTERNAL TANK

### I. BACKGROUND

The External Tank appears to be simple in concept. The liquid oxygen and hydrogen tanks are basically of a conventional design. However, the Tank has turned out to have significant engineering and manufacturing challenges. There are also the challenges of designing the fore and aft Orbiter attachment hardware, the external insulation and lightning protection systems. Thus a Panel member was assigned to this important area.

Information on the status of the External Tank has been obtained through formal presentations at JSC and Rockwell International and through detailed review of the system at MSFC. Also, a visit was made to Martin-Marietta at Michoud earlier. In addition, a study was made of the Hazards Analysis Report, MMC-ET-RA01-A, dated October 17, 1975.

### II. OBSERVATIONS

The hazard status summarized in October 1975 was:

- A. 58 hazards identified.
- B. 31 hazards submitted to NASA for evaluation.
- C. 2 residual hazards proposed for acceptance as continuing hazards by NASA.
- D. 25 hazards resolved.

At the Quarterly Review on May 6, 1976, the list of hazards was revised to show the following changes:

- A. 67 hazards identified.
- B. 33 hazards submitted to NASA for evaluation.
- C. 2 residual hazards proposed for acceptance as continuing hazards by NASA.
- D. 32 hazards resolved.

It would be premature of the Panel to comment on the detail deliberations among the contractors and the NASA Centers until firm decisions have been reached. It should be pointed out, however, that the classification above of "Residual Hazards" corresponds to the concept of a "Risk List" as suggested in 1975 by the Hawkins Committee for the entire Shuttle system. The Panel concurs in the concept that such a list should be the prime focus for reviewing the readiness for operation of a subsystem of the Shuttle such as the External Tank and commends the Shuttle management and Marshall for this method of monitoring the hazards inherent in the system.

Several hazards described in the above-referenced report should be addressed in subsequent studies.

A. The breakdown of the hazards into the functional list selected caused a great deal of cross referencing. Some other breakdown might make a review by outsiders simpler and more productive.

B. The problem of flammability of the Thermal Protection System in the presence of gaseous or liquid propellants suggests that a com-

plete review of propellant leakage and possible spillage may be of value. The toxicity of the polyurethane foam with a flame retardant needs more study and a systems decision. The addition of the flame retardant makes the residual ash and the gas emission more objectionable, perhaps unacceptable, if a fire should occur. A fire may be avoidable and unlikely, but if one should occur, the questionable improvement of a fire retardant makes the insulation material in use more dangerous. The effectiveness of the retardant in case of an oxygen leak is questionable. There is the additional fact that the external, or bonding, insulation of the External Tank is temperature sensitive. Any lengthy exposure to direct solar heating might degrade the integrity of the Thermal Protection System (CPR 421).

C. There was no discernable reference in the reports to previous NASA or contractor experience on launch vehicles which must have been subject to similar fire hazards. Solutions which were reached on such vehicles must be equally applicable to the External Tank and would be far more convincing to reviewers than some of the test programs or explanations which were offered to mitigate or remove the hazard.

D. A series of lightning tests performed recently showed that the protection system problem is not yet solved; specifically, the bonding of multiple spray-on paint strips to a single path solid metal in the form of the vent line. In addition, the selection of

the proper spray-on conduction paint itself needs more test and studies.

E. The occurrence of geysering during filling of the long suction lines has to be thoroughly tested, and the baffles inside the tank must be protected. Tests are still forthcoming.

F. Large cryogenic separation fittings subject to water and nitrogen icing might be troublesome to guarantee a proper disconnect. To date, no ground separation test (even simulated) is planned.

### III. ASSESSMENT AND RECOMMENDATIONS

It is the opinion of the Panel member who reviewed the External Tank status, that there are no insurmountable risks that cannot be adequately controlled for safe operations. It is suggested that the Panel participate through its individual members, in subsequent critical design or normally scheduled reviews and that the entire Panel be exposed to the final "Residual Hazards" which the program managers believe should be accepted for first orbital flight and subsequent operations.

A. The target performance data of the orbiter systems were quoted and finalized as a point in time when finalized loads, aerodynamic, thermodynamic, vibration, and vibro-acoustic, were in a preliminary state. Weights and propellants have only minor allowances for variations. Finalized data in all environmental fields will not be available until late in the test program and may result in a costly

redesign and, sooner or later, performance variations may well result.

B. Critical mechanical activities like the complex separation of the External Tank and Orbiter will be experienced for the first time under environmental conditions during the first orbital flight. If at all possible, it would be prudent to include an environmental separation ground test in the program. A flight failure can neither be observed nor measured and could well lead to a total loss of the Orbiter.

C. A reasonable consistency in the quality of the External Tank in order to achieve maximum reliability and safety of the manned flight is best assured by continuing production. Shutdown and the subsequent reopening of the production line will interrupt the learning curve and compromise a reasonable, low price of the throw-away External Tank which is best achieved by an acceptable continuous production rate. The actual use of the External Tank is governed by entirely different aspects. A launch delay, weather, mechanical difficulties, payload availability, or other unpredictable events, will create a possible storage problem for the External Tank. It would be advisable to assure suitable limited storage space for these large External Tanks. Storage conditions would have to be controlled to insure against degradation.

D. Lightning tests have shown some weaknesses of the test specimens representing the intended External Tank design. It is suggested

that a "Lightning Protection Committee," or "**Study Group**," approve the finalized lightning protection measures, not only for the launch pad, but for the vehicle in flight as well. These reviews should include proper bonding and prevention of static charges.



10.0 SOLID ROCKET BOOSTER

Dr. William A. Mrazek



## 10.0 SOLID ROCKET BOOSTER

### I. BACKGROUND

The technology of large solid rockets is well developed, and many operational units have been found to be reliable and trouble-free. The Panel recognizes the importance of this element and the need for high reliability. The development program on this element is now reaching the stage for more intensive review.

Several Solid Rocket Booster Quarterly Reviews were attended and, in addition, insight was gained by visits with the project management staff. Up to this date, contractor visits have not been made because of the early status of the project. The last contract for the assembly of the booster is about to be let as of the date of this report.

Nevertheless, the latest issue of the JSC Report #09990A published March 8, 1976, titled "Major Safety Concerns of Space Shuttle Program" lists two open safety concerns, INTG-11 and INTG-12, pertaining to the Solid Rocket Booster.

INTG-11 - "A Nozzle Extension Separation Failure" will be disposed of prior to the first launch.

INTG-12 - "Ignition Overpressure" Completion of a comprehensive study is scheduled for July 1976. It is evident that late adverse study results might have a considerable impact on cost, performance, and schedule.

## II. OBSERVATIONS

Despite the diligent application of available experience and data, the project recognizes major uncertainties in design criteria. Lift-off loads, thermal environment and changes will have an impact on cost, schedule, and performance. Twelve concerns were recognized by project management and discussed in detail. To obtain a conclusive picture of the progress made, it was suggested by the Panel members that at following reviews, the status of the above concerns, as well as others, be monitored.

## III. ASSESSMENT AND RECOMMENDATIONS

A. The auxiliary power unit supplying oil pressure to the actuators of the boosters uses as its prime mover a hydrazine-driven turbine to operate the pumps. The exhaust stacks of all four units located in both boosters allow the entry of sea water into the catalyst bed of the fuel system after splashdown. To date eleven (11) mission duty cycle tests of the unit have been completed during which the catalyst bed was exposed to salt water for ten (10) hours each cycle. After retrieval from the water, the bed was flushed out and successfully fired in all cases. The "reconditioning" system must assure adequate flushing is accomplished after each and every salt water exposure.

B. A molded fiber-reinforced plastic cover of adequate strength could be designed and produced to enclose the entire APU for protection against sea water dunking. The savings in the long run could easily offset the initial cost.

The Panel will be devoting increased attention to the Solid

Rocket Booster system during the year ahead. Hazards associated with Shuttle system assembly in the VAB at KSC will be included in such surveillance.